

Serial No. Not Yet Assigned

Atty. Doc. No. 2002P06704US

Amendments To the Claims:

Please amend the claims as shown.

1. (currently amended) A Ggas turbine (1) with comprising:

a compressor for compressing air;

an annular combustion chamber (4) operatively connected to the compressor, the combustion chamber having a combustion area (24) of which is bounded by an annular combustion chamber outer wall (26) on the one hand and an annular combustion chamber inner wall (28) located therein on the other hand, wherein the combustion chamber inner wall (28) is formed by a plurality of wall elements attached to a support structure of the combustion chamber inner wall (28), and wherein the support structure is formed by a plurality of sub-components (30) abutting each other at a horizontal parting joint, said the sub-components being connected to each other in the area of the parting joint via a plurality of screw connections (32) oriented at an angle to the inner wall surface; and

an airfoil section operatively connected to the combustion chamber.

2. (currently amended) A Ggas turbine (1) according to Claim 1, wherein a key (34) is assigned to the or each at least one screw connection (32).

3. (currently amended) A Ggas turbine (1) according to Claim 1 or 2, wherein the combustion chamber outer wall (26) of the annular combustion chamber (4) is implemented formed in two parts and formed by a lower part (38) interacting with an upper part (36).

4. (currently amended) A Ggas turbine (1) according to one of Claims 1 to 3, wherein the combustion chamber inner wall (28) and/or the combustion chamber outer wall (26) is fitted with a lining formed by a plurality of heat shield elements (40).

5. (currently amended) A Ggas turbine (1) according to Claim 4, wherein the heat shield elements (40) are attached to the combustion chamber inner wall (28) or the combustion chamber outer wall (26) by means of a tongue and groove system.

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6. (new) A gas turbine according to Claim 2, wherein the outer wall is formed in two parts.
7. (new) A gas turbine according to Claim 2, wherein the inner wall and/or the outer wall is fitted with a lining formed by a plurality of heat shield elements.
8. (new) A gas turbine according to Claim 3, wherein the inner wall and/or the outer wall is fitted with a lining formed by a plurality of heat shield elements.
9. (new) A gas turbine according to Claim 7, wherein the heat shield elements are attached to the inner wall or the outer wall by means of a tongue and groove system.
10. (new) A gas turbine according to Claim 8, wherein the heat shield elements are attached to the inner wall or the outer wall by means of a tongue and groove system.
11. (new) A gas turbine according to Claim 1, wherein the combustion chamber is an annular combustion chamber.
12. (new) A gas turbine according to Claim 1, wherein the sub-components abutting each other.
13. (new) A gas turbine according to Claim 1, wherein the airfoil section is operatively adapted to turn a shaft.
14. (new) A gas turbine according to Claim 1, wherein the airfoil section is operatively adapted to drive the compressor or a generator.
15. (new) A gas turbine according to Claim 3, wherein a lower part interacts with an upper part.
16. (new) A gas turbine according to Claim 6, wherein a lower part interacts with an upper part.

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17. (new) A combustion chamber comprising:

 a plurality of burners to burn a fuel;

 an outer wall;

 an inner wall; and

 a combustion area bounded by the outer wall and the inner wall, the inner wall formed by a plurality of wall elements attached to a support structure of the inner wall, and the support structure formed by a plurality of abutting sub-components, the sub-components connected to each other in the area of a parting joint via a plurality of screw connections oriented at an angle to the inner wall surface.

18. (new) A combustion chamber according to Claim 17, wherein the combustion chamber is an annular combustion chamber.